Application No. 10/628,556
Amendment dated October 1, 2007
Reply to Office Action of May 1, 2007

Amendments to the Claims:

This listing of claims will replace all prior versions and listings of claims in the application.

Listing of Claims

1. (Currently Amended) A casing for a turbofan engine, substantially encasing at least a fan assembly, a compressor assembly, a combustor assembly and a turbine assembly, the casing comprising:

a fan case portion surrounding the fan assembly;

an intermediate case portion projecting axially rearwardly from the fan case portion; and

a gas generator case portion extending axially rearwardly from the intermediate case portion and housing the combustor assembly,

wherein the fan case portion, the intermediate case portion and the gas generator case portion are integrally joined together, thereby forming an integral casing.

2. (Original) The casing as claimed in claim 1 wherein the fan case portion, the intermediate case portion and the gas generator case portion are made of the same material.

3. (Original) The casing as claimed in claim 1 wherein the intermediate case portion further comprises an integral compressor shroud portion and an integral bearing mount portion.

4. (Cancelled)

5. (Currently Amended) The casing as claimed in claim 1 wherein a first weld joins the ~~individual~~ fan case portion~~,~~ to the intermediate case portion, and wherein a second weld joins the gas generator case portion ~~are fabricated individually and welded together~~to the intermediate case portion.

6. (Original) The casing as claimed in claim 1 wherein the fan case portion, the intermediate case portion and the gas generator case portion are joined together by flangeless connections.

7. (Original) A bypass turbofan engine comprising:

at least a fan, a compressor, and a gas generator disposed in flow series within the engine, and a bypass airflow defined around at least the compressor and gas generator; and

a one-piece casing substantially encasing the fan, compressor and gas generator.

8. (Original) A turbofan engine as claimed in claim 7 wherein the casing further comprises an integral compressor shroud encircling blade tips of the compressor.

9. (Cancelled)

10. (Original) A turbofan engine as claimed in claim 9 wherein bearing seat is configured to provide integral damping to the compressor shaft bearing.

11. (Original) The turbofan engine as claimed in claim 7 wherein the casing at least partially defines a by-pass air flow passage within the engine.

12. (Original) A turbofan engine for an aircraft comprising:

a rotating assembly including a propulsive fan portion, a compressor portion, and a gas generator portion, the rotating assembly having an axial length; and

a generally tubular casing assembly enveloping the rotating assembly substantially along the axial length thereof and thereby defining a main flow path through the engine, wherein the casing assembly is an integrated single piece.

13. (Original) The turbofan engine for aircraft as claimed in claim 12 wherein the casing assembly further comprises a integral shroud section encircling a plurality of compressor blade tips of the compressor portion.

14. (Original) The turbofan engine for aircraft as claimed in claim 12 wherein the casing assembly further comprises an integral bearing seat for operatively supporting a compressor shaft of the compressor portion.

15. (Original) The turbofan engine for aircraft as claimed in claim 12 wherein the casing defines at least a portion of a by-pass air duct of the engine.

16. (New) The bypass turbofan engine as claimed in claim 7, wherein the one-piece casing defines an internal space, and wherein the fan, the compressor and the gas generator are insertable and removable from said internal space without opening said one-piece casing.

17. (New) The casing as claimed in claim 1, wherein the intermediate case portion comprises a splitter including inner and outer annular walls extending axially rearwardly from a leading edge tip, the outer annular wall forming a radially inner boundary of a bypass air flow, the inner annular wall forming a radially outer boundary of a main gas path of the turbofan engine.

18. (New) The casing as claimed in claim 1, wherein the fan case portion has an inner shroud which is integrally connected to an inner hub of the intermediate case portion.